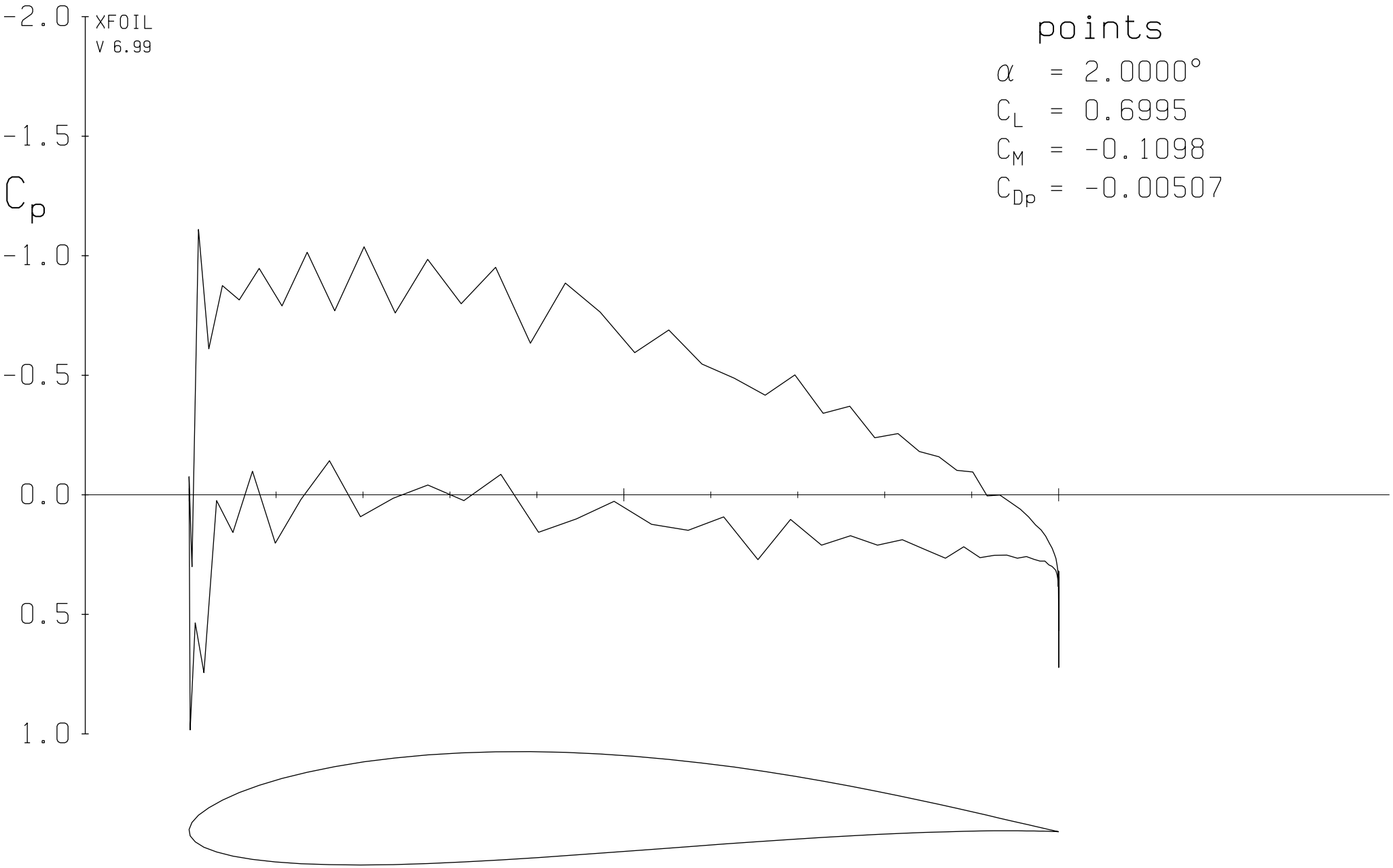
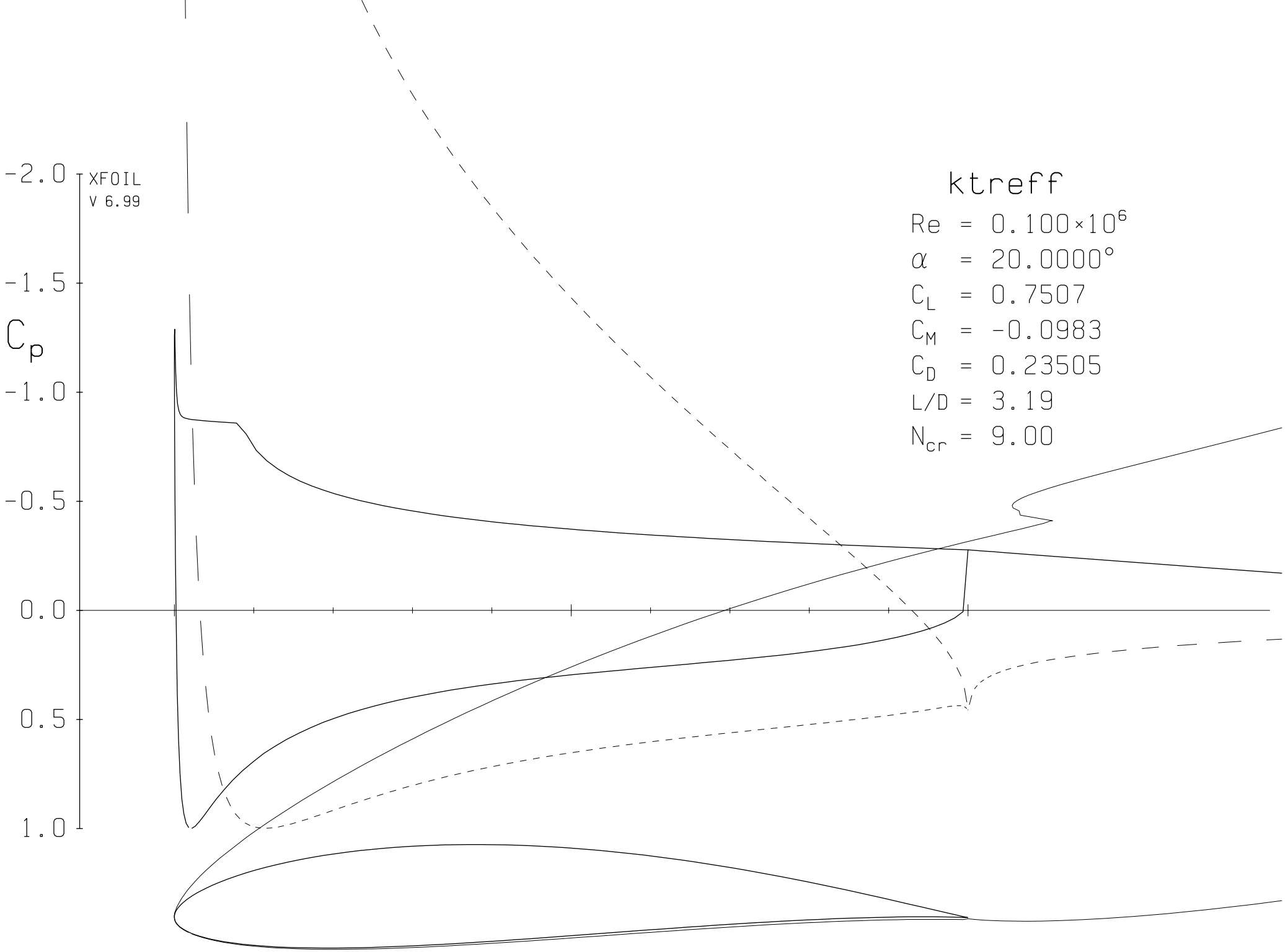
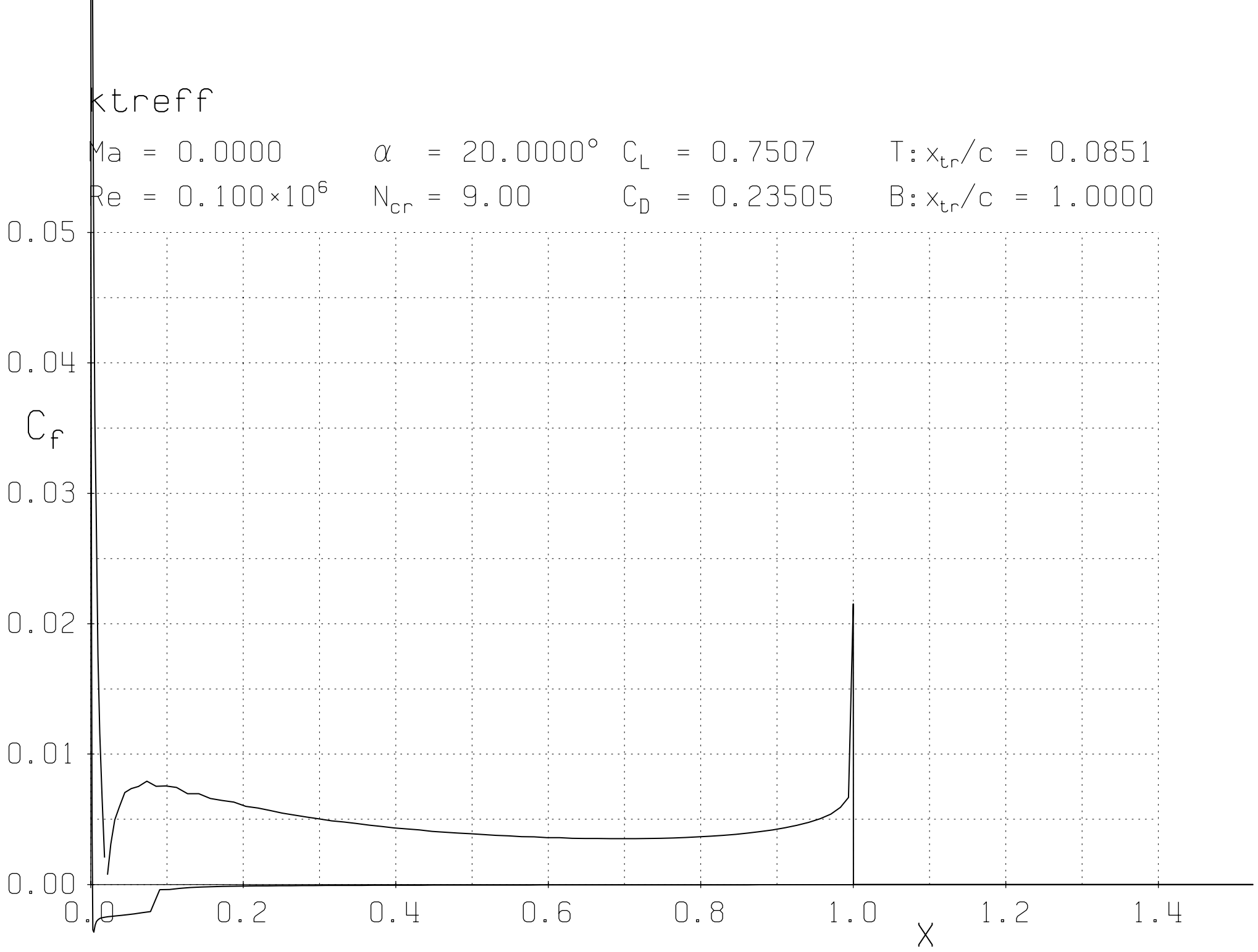
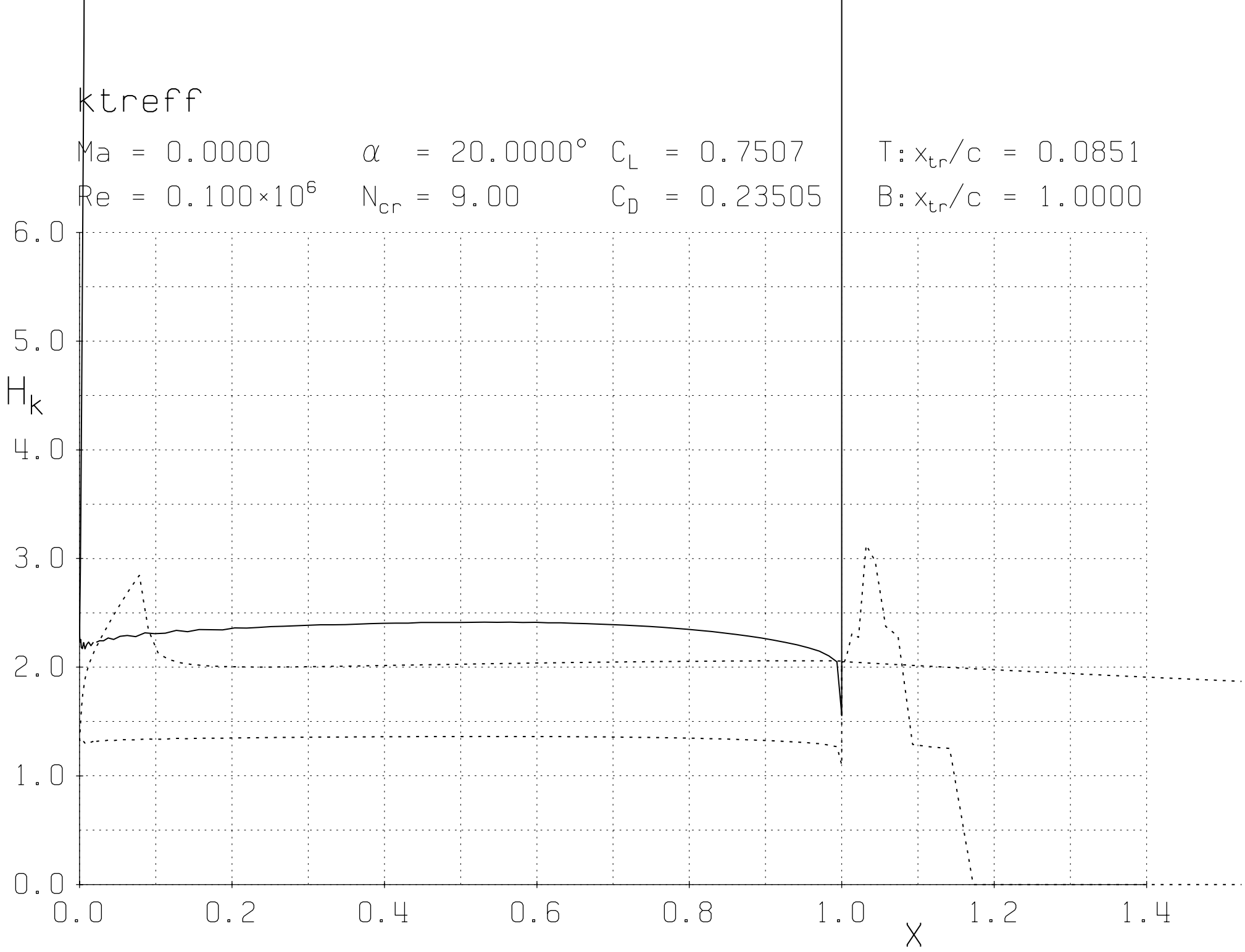
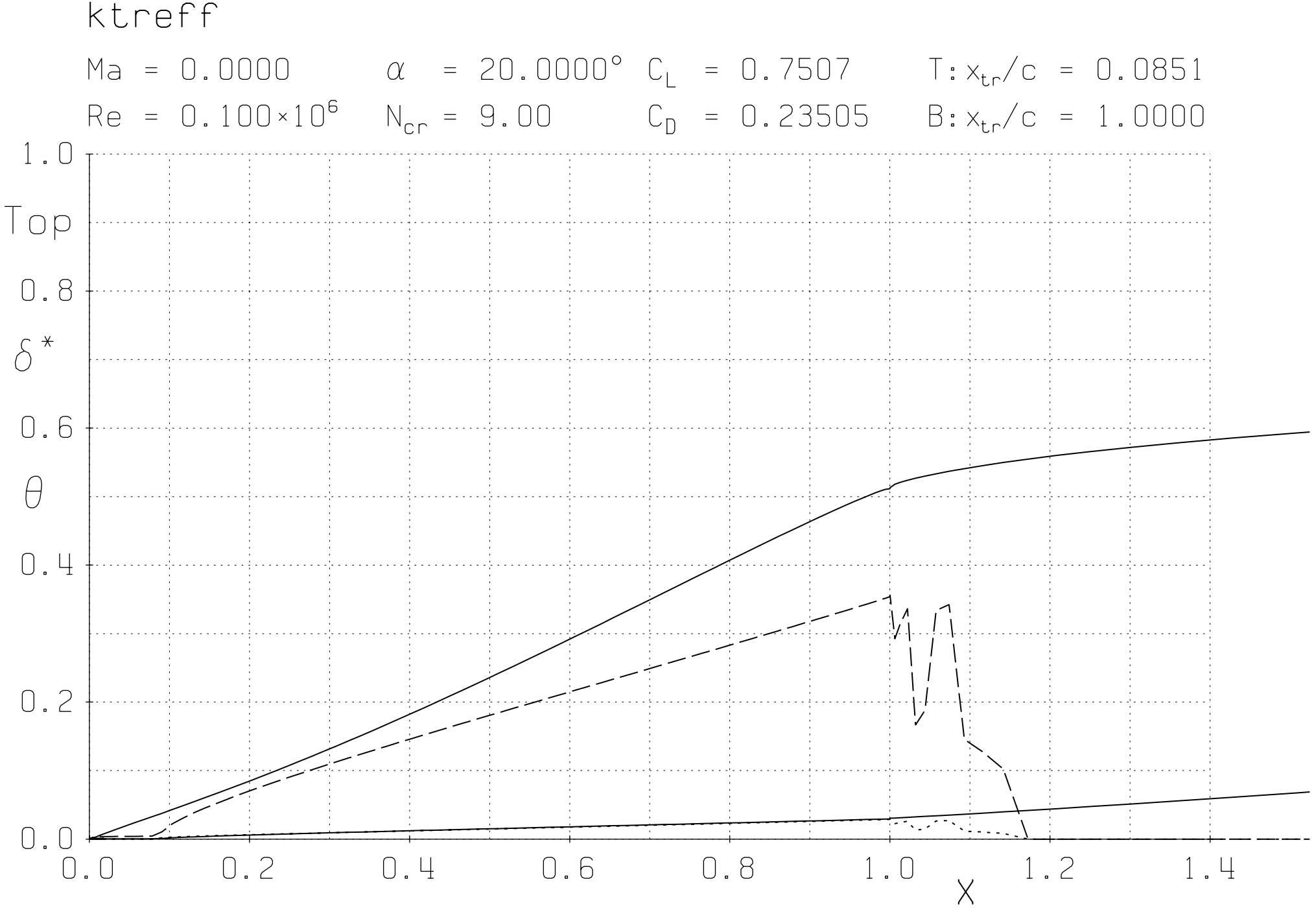
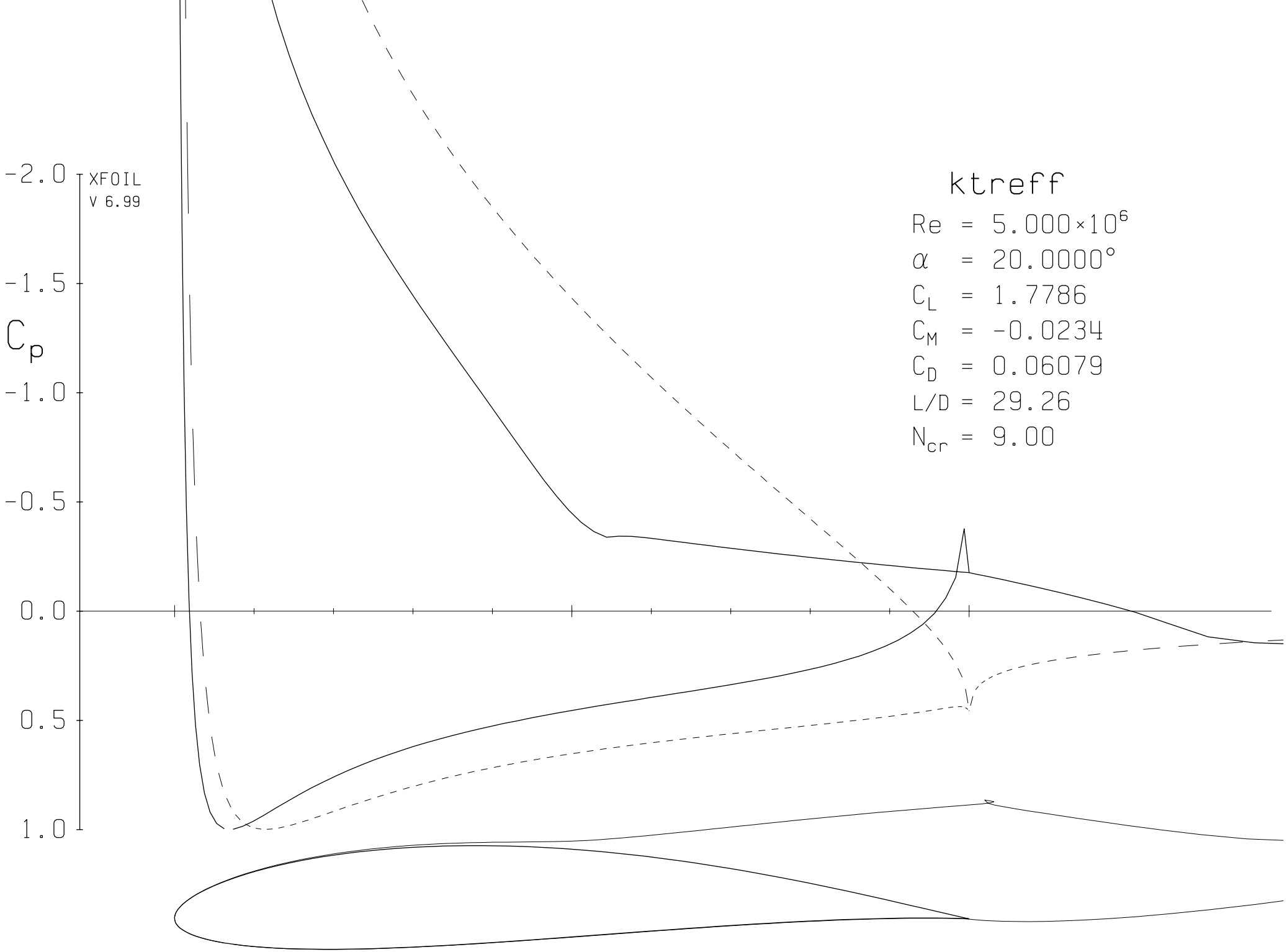

XFOIL
V 6.99
points
α = 2.0000°
C_L = 0.6995
C_M = -0.1098
C_Dp = -0.00507
C_p
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0

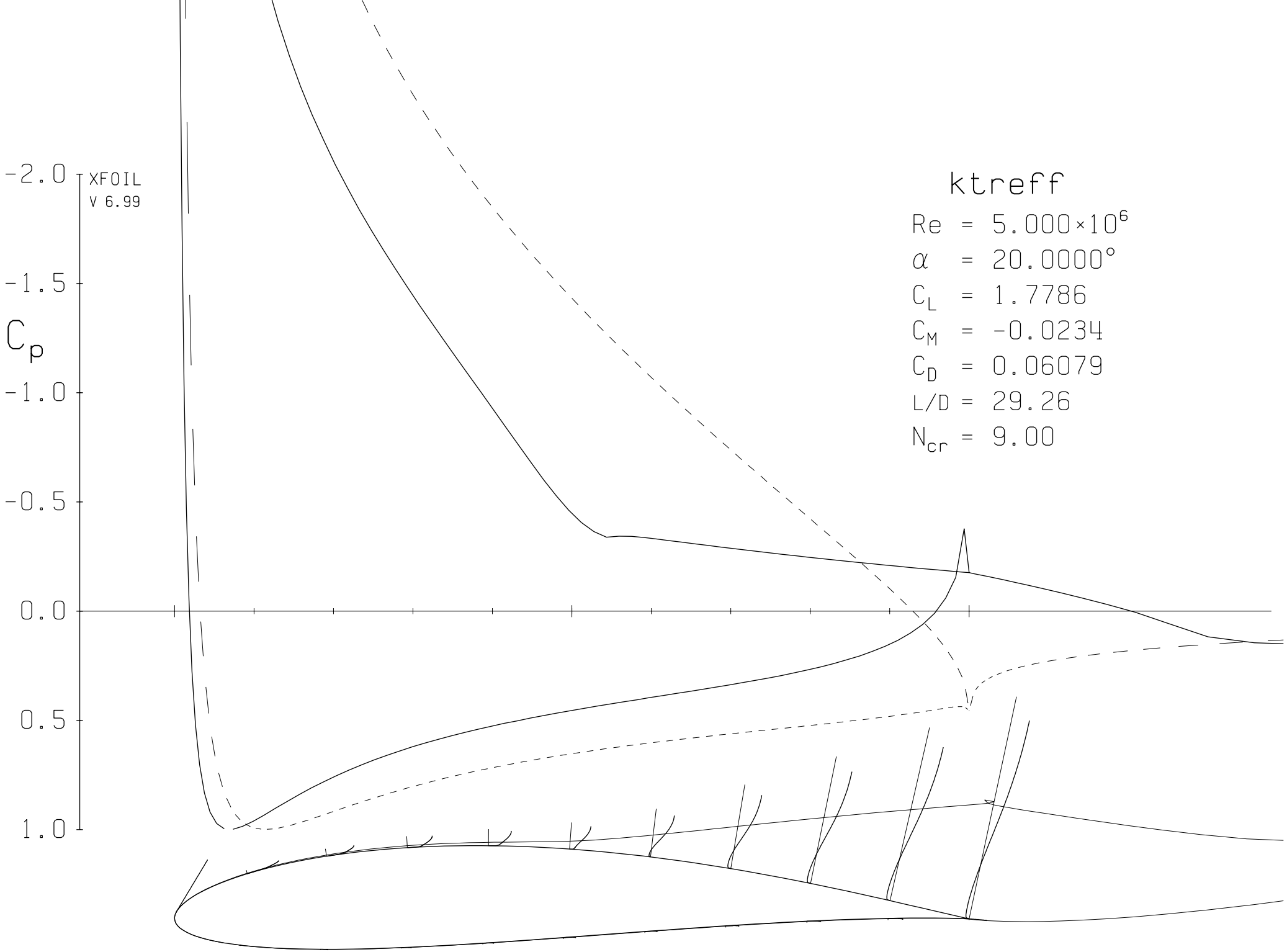

XFOIL
V 6.99
ktreff
Re = 0.100×10^6
α = 20.0000°
C_L = 0.7507
C_M = -0.0983
C_D = 0.23505
L/D = 3.19
N_cr = 9.00
C_p
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0

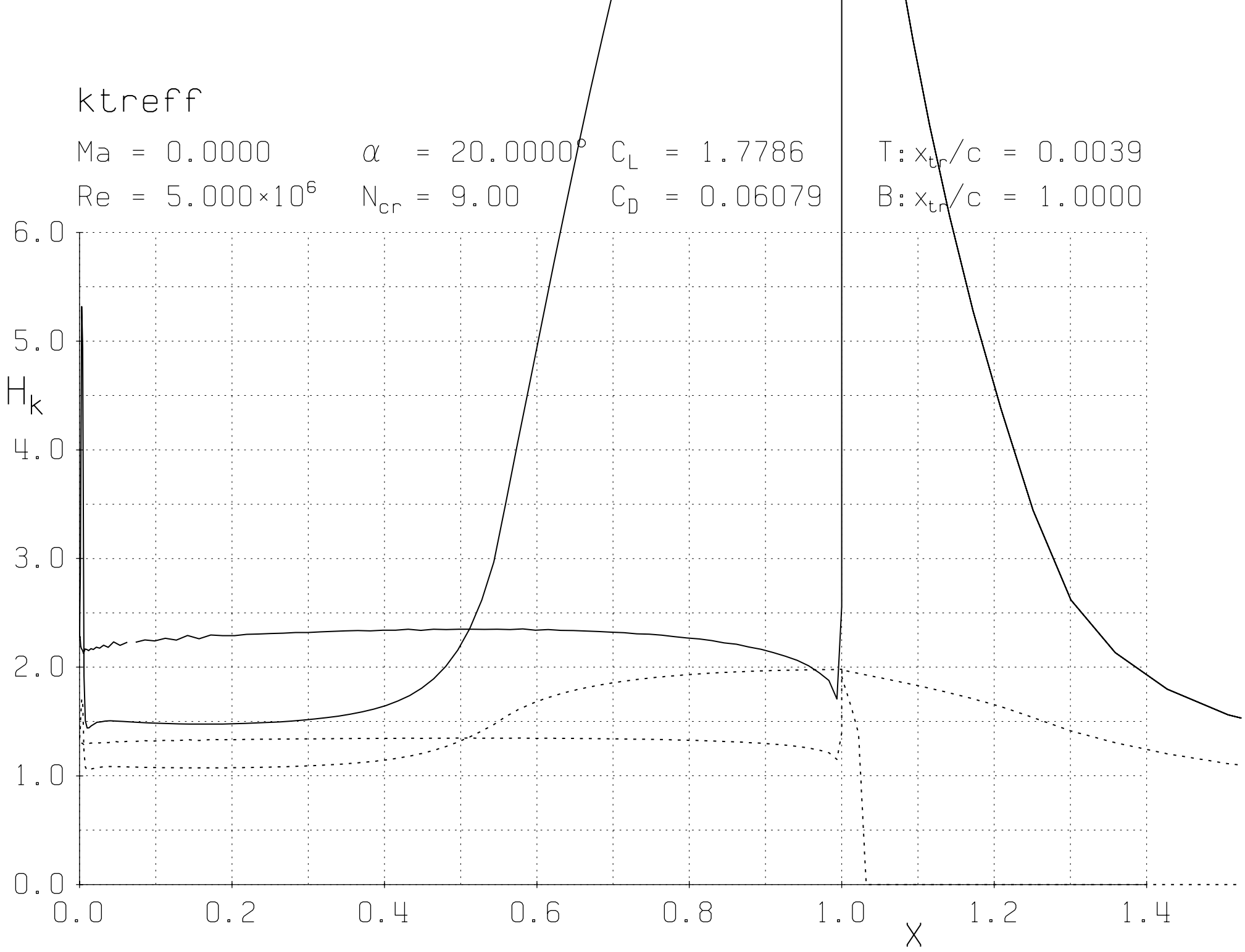

ktreff
Ma = 0.0000 α = 20.0000° C_L = 0.7507 T: x_tr/c = 0.0851
Re = 0.100×10^6 N_cr = 9.00 C_D = 0.23505 B: x_tr/c = 1.0000
0.05
0.04
C_f
0.03
0.02
0.01
0.00
0.0
0.2
0.4
0.6
0.8
1.0
1.2
1.4
X

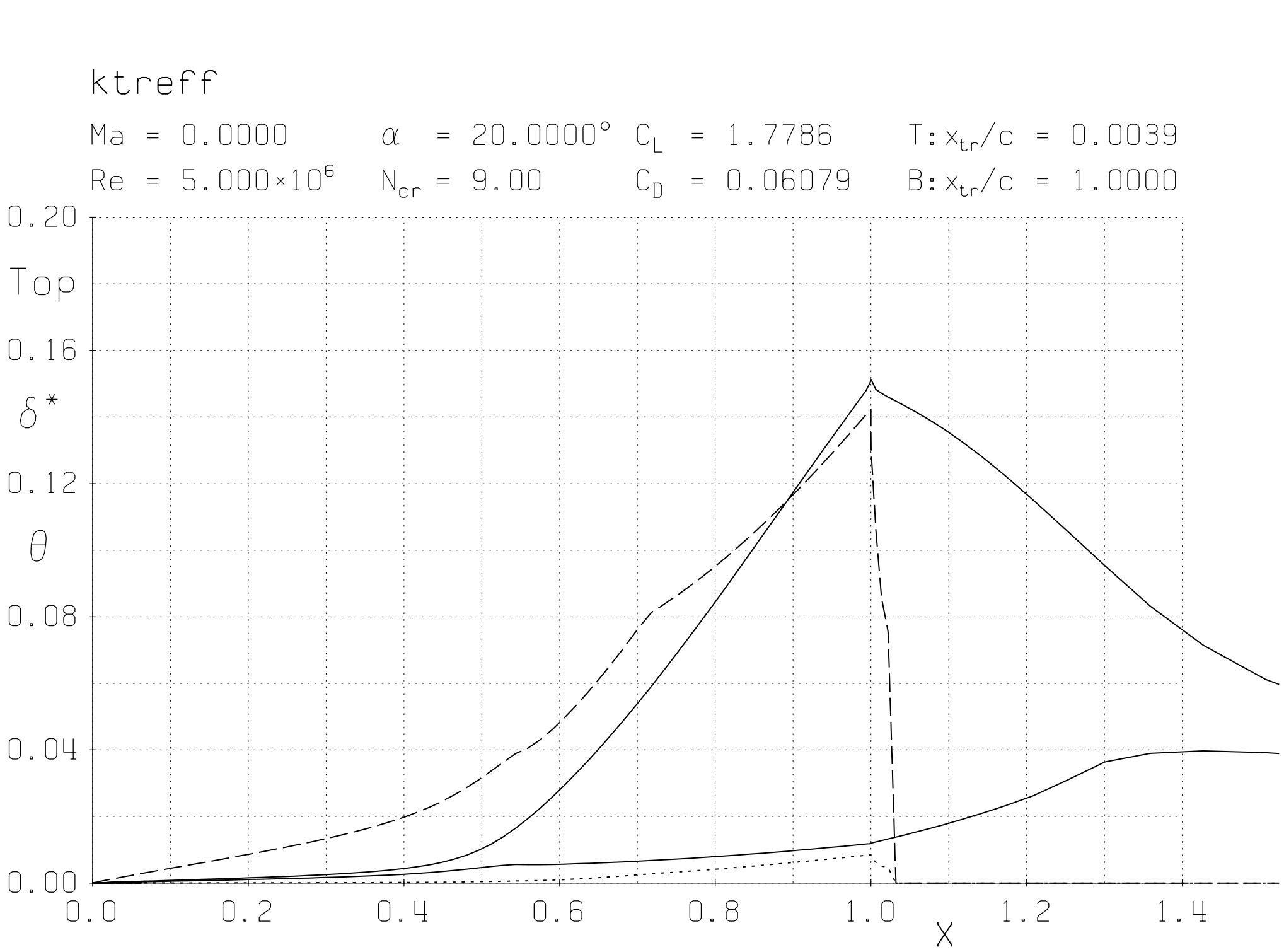

ktreff
Ma = 0.0000
α = 20.0000°
C_L = 0.7507
T: x_tr/c = 0.0851
Re = 0.100×10^6
N_cr = 9.00
C_D = 0.23505
B: x_tr/c = 1.0000
H_k
6.0
5.0
4.0
3.0
2.0
1.0
0.0
0.0
0.2
0.4
0.6
0.8
1.0
1.2
1.4
X

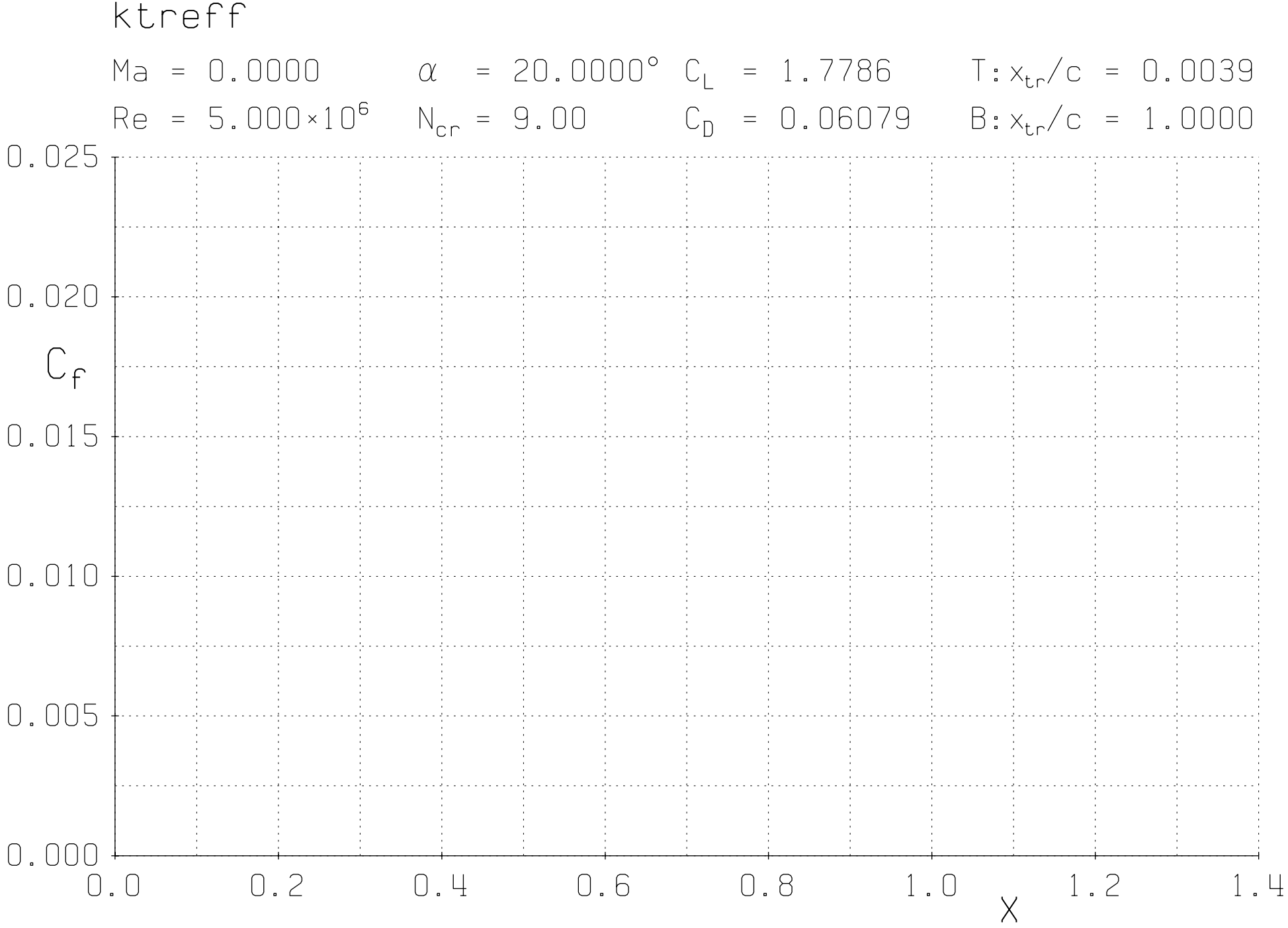

ktreff
Ma = 0.0000 α = 20.0000° C_L = 0.7507 T: x_tr/c = 0.0851
Re = 0.100×10^6 N_cr = 9.00 C_D = 0.23505 B: x_tr/c = 1.0000
Top
δ*
θ
1.0
0.8
0.6
0.4
0.2
0.0
0.0
0.2
0.4
0.6
0.8
1.0
1.2
1.4
X


XFOIL
V 6.99
ktreff
Re = 5.000×10^6
α = 20.0000°
C_L = 1.7786
C_M = -0.0234
C_D = 0.06079
L/D = 29.26
N_cr = 9.00
C_p
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0


XFOIL
V 6.99
ktreff
Re = 5.000×10^6
α = 20.0000°
C_L = 1.7786
C_M = -0.0234
C_D = 0.06079
L/D = 29.26
N_cr = 9.00
C_p
-2.0
-1.5
-1.0
-0.5
0.0
0.5
1.0


ktreff
Ma = 0.0000 α = 20.0000° C_L = 1.7786 T: x_tr/c = 0.0039
Re = 5.000×10^6 N_cr = 9.00 C_D = 0.06079 B: x_tr/c = 1.0000
6.0
5.0
4.0
3.0
2.0
1.0
0.0
H_k
0.0
0.2
0.4
0.6
0.8
1.0
1.2
1.4
X


ktreff
Ma = 0.0000 α = 20.0000° C_L = 1.7786 T: x_tr/c = 0.0039
Re = 5.000×10^6 N_cr = 9.00 C_D = 0.06079 B: x_tr/c = 1.0000
0.20
Top
0.16
δ*
0.12
θ
0.08
0.04
0.00
0.0
0.2
0.4
0.6
0.8
1.0
1.2
1.4
X


ktreff
Ma = 0.0000 α = 20.0000° C_L = 1.7786 T: x_tr/c = 0.0039
Re = 5.000×10^6 N_cr = 9.00 C_D = 0.06079 B: x_tr/c = 1.0000
0.025
0.020
C_f
0.015
0.010
0.005
0.000
0.0
0.2
0.4
0.6
0.8
1.0
1.2
1.4
X